

Comparison of numerical and experimental results over a NACA0025 airfoil undergoing separation

Abstract

This paper examines unsteady numerical simulation of a three-dimensional flow over two different symmetric NACA airfoils. The airfoils are at various angles of attack and various low Reynolds numbers ($5 \times 10^4 \sim 5 \times 10^6$). The Spalart-Allmaras, and $k-\epsilon$ models, as well as LES and E-LES approaches are used and compared to experimental results. The capability of each turbulent model to capture characteristics at low Reynolds number is discussed. The main objective in this study is to capture the behavior of boundary-layer separation with respect to experiments at critical Reynolds numbers and then to establish a useful solution methodology describing transition in the boundary layer of airfoils accurately. Fully attached flow, boundary-layer separation, and boundary-layer with reattachment conditions are studied. Although LES and E-LES provide the highest computation cost, results show the accuracy of these methods at low Reynolds numbers ($< 150 \times 10^3$). For higher Reynolds number, turbulence models provide a fair agreement with experiments; while the processing cost of LES method is very high. Finally, the impact of airfoil thickness, Reynolds number, and angle of attack on the boundary-layer separation and consequently airfoil performance are studied in detail.

Keywords: reynolds numbers, strouhal number, spalart-allmaras model, k- ϵ model, eppler 387 airfoil, NACA 0025 airfoil, smagorinsky

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Abbreviations: SEP, separation; MAX, maximum; LES, large eddy simulation; E-LES, embedded large eddy simulation; SAS, scale-adaptive simulation.

Introduction

The performance of airfoils operating at low relative wind speeds (low Reynolds number) includes applications such as wind turbines, remotely piloted vehicles, sailplanes, human powered vehicles and high altitude devices.¹⁻³ At lower Reynolds numbers, both lift and drag characteristics are different from high Reynolds number flows.

Lian et al.,² studied unsteady boundary-layer separation from an Eppler 387 airfoil at low Reynolds number numerically and compared their results with wind tunnel measurements. They found unsteady large-scale structure might control the low-Reynolds-number separation bubble reattachment with small-scale turbulence playing a secondary role. Lin JCM,⁴ Drela,⁵ Drela et al.,⁶ used a fully coupled nonlinear system and predicted transitioning separation bubbles and their associated losses.

At low Reynolds numbers, flow separation occurs on the airfoils at small angles of attack, between 2° and 6° , with noticeable effects on airfoil efficiency. Moreover, with decreased Reynolds number, the length of the separation region decreased. When the maximum camber position was moved toward the trailing edge, the flow separation length was substantially reduced.⁷ Adding a burst control plate to increase the turbulent kinetic energy was found to reattach the shear layer to the airfoil surface.⁸

Measurements have shown that the boundary-layer separates downstream of an adverse pressure gradient, leading to the formation of a thin separation bubble and the second formed vortex loop retains

its identity in the turbulent boundary layer and persists until the end of the test section.⁹ Experiments were performed for a range of low Reynolds numbers (50×10^3 – 200×10^3) and various angles of attack in a wind tunnel and over the NACA 0025 airfoil with a chord length c of 0.3m and a span of 0.88m. Boundary layer separation occurred over a large region on the upper surface of the airfoil for the entire range of angles of attack at $Re=50 \times 10^3$ and $Re=100 \times 10^3$.^{10,11} The results suggested that higher amplitude excitations might affect the airfoil performance at lower Reynolds numbers.¹² In addition, it was pointed out that the magnitude of the acoustic excitation influence on the wake structure correlates with the extent of the improvement in the airfoil performance.

In this work, the boundary-layer behaviors of the symmetric airfoil NACA 0025 are studied numerically and are compared with experiments. The goal of this study is to find the most appropriate turbulence approaches for the simulation of boundary layer separation at different ranges of Reynolds number by comparing the Spalart-Allmaras and $k-\epsilon$ models and LES methods with experimental data.¹⁰

Governing equations and computational models

Turbulent models

Spalart and allmaras model: The Spalart and Allmaras¹³ model is a modified turbulence model for aerospace applications involving wall-bounded flows. This has been found to have good performance for boundary layers subjected to adverse pressure gradients.¹⁴ In its original form, it is effectively a low-Reynolds number model, requiring the viscous-affected region of the boundary layer to be properly resolved and does not have non-local components in the equation, e.g. y^+ . The

transported variable $\tilde{\nu}$ is identical to the turbulent kinematic viscosity except in the viscous affected region,

$$\rho \frac{D\tilde{\nu}}{Dt} = \rho c_{b1} \tilde{S}\tilde{\nu} + \frac{1}{\sigma_{\tilde{\nu}}} \left[\frac{\partial}{\partial x_j} \left\{ \left(\mu + \rho \tilde{\nu} \right) \frac{\partial \tilde{\nu}}{\partial x_j} \right\} + \rho c_{b2} \left(\frac{\partial \tilde{\nu}}{\partial x_j} \right)^2 \right] - \rho c_{w1} f_w \frac{\tilde{\nu}}{d^2} \quad (1)$$

Where $\rho c_{b1} \tilde{S}\tilde{\nu}$ and $\rho c_{w1} f_w \tilde{\nu}/d^2$ are, the production and destruction of turbulent viscosity respectively that occurs in the near-wall region. And, $\sigma_{\tilde{\nu}}$ and C_{b2} are user-determined constants. The modified turbulent kinematic viscosity is set to zero at walls. When the mesh is generated appropriately to solve the laminar sub-layer, the wall shear stress is obtained from the laminar stress-strain equation.¹⁵

$$\frac{u}{u_{\tau}} = \frac{\rho u_{\tau} y}{\mu} \quad (2)$$

The $k - \epsilon$ – two-equation model: The two-equation models solve two separate transport equations and allow the turbulent velocity and length scales to be independently determined.¹⁶ The realizable $k - \epsilon$ model was proposed by Shih et al.¹⁷ and was intended to solve some deficiencies of the traditional $k - \epsilon$ models by adopting a new eddy-viscosity formula and a new dissipation equation, based on the dynamic equation of the mean-square vorticity fluctuation,

$$\rho \frac{Dk}{Dt} = \frac{\partial}{\partial x_j} \left[\left(\mu + \frac{\mu_t}{\sigma_k} \right) \frac{\partial k}{\partial x_j} \right] + G_k + G_b - \rho \epsilon - Y_M \quad (3)$$

$$\rho \frac{D\epsilon}{Dt} = \frac{\partial}{\partial x_j} \left[\left(\mu + \frac{\mu_t}{\sigma_{\epsilon}} \right) \frac{\partial \epsilon}{\partial x_j} \right] + \rho c_1 S \epsilon - \rho c_2 \frac{\epsilon^2}{k + \sqrt{\nu \epsilon}} + c_{1\epsilon} \frac{\epsilon}{k} c_{3\epsilon} G_b \quad (4)$$

where, $\sigma_k, \sigma_{\epsilon}, C_{1\epsilon}, C_{2\epsilon}$ are empirical constants for $k - \epsilon$ model. $\eta = Sk / \epsilon$. G_k and G_b represent the generation of turbulent kinetic energy due to mean velocity gradients and buoyancy, respectively. Y_M is the contribution of the fluctuating dilatation in the compressible turbulence to the overall dissipation rate.¹⁸ In this model, the eddy viscosity is calculated from Eq. (6) with variable C_{μ} .^{4,15}

$$C_{\mu} = \frac{1}{A_0 + A_s \frac{U^*}{\epsilon}} \quad (5)$$

A_0, A_s , and U^* are functions of velocity gradients.

$$\mu_t = \rho C_{\mu} \frac{k^2}{\epsilon} \quad (6)$$

It is a semi-empirical model.¹⁶

To model transition from laminar to turbulent flow in this aerodynamic simulation, the approach of Eleni et al was used.¹⁶ The problem was simulated in Fluent after defining a laminar region (close to leading edge) and a turbulent zone. If the C_D simulation result was greater than that of the experiment, the transition point moves closer to the trailing edge. This an iterative procedure until an accurate C_D ^{16,18} is obtained.

Large eddy simulation (LES) and embedded large eddy simulation (E-LES): LES still requires substantially finer meshes than those typically used for RANS calculations. In addition, LES has

to be run for a sufficiently long flow-time to obtain stable statistics of the flow being modeled. As a result, the computational cost involved with LES is normally considerably higher than that for steady RANS calculations in terms of memory and CPU time.^{19,20}

LES has had a limited impact on industrial CFD simulations, mainly because of its high computational costs. There are few technical applications where LES can be applied within the entire computational domain. Such flows are typically of low Reynolds numbers or with unimportant wall boundary layers. The high-resolution requirements for wall-bounded flows even at moderate Reynolds numbers have, in particular, severely limited the usage of LES.^{15,21}

In order to allow the resolution of large turbulent structures in industrial flow simulations, hybrid models like Scale-Adaptive Simulation (SAS) have been employed. In such situations, zonal models are desirable, where a clear distinction between RANS and LES regions can be made, and where turbulence is converted from RANS to LES by suitable methods at the interface. One such approach is E-LES, where there are separate RANS and LES zones produced during the grid generation phase, then appropriate models for each zone are chosen and define the appropriate treatment at the interface. Usually, all RANS models can be used in the RANS region except for the Spalart-Allmaras one-equation model as it cannot provide the required turbulent length scale at the interface. Hence, for the RANS zone of the E-LES approach, realizable $k - \epsilon$ is used. In this model, the smaller scales and their interaction with the resolved scales have to be modeled through the sub grid scale term. The high-pass filtered Smagorinsky (HPF) eddy-viscosity model has been used here with has developed independently by Vreman (2003) and is closely related to the variational multiscale approach. It is known that by introducing a high-pass filtering the classical Smagorinsky closure could be applied without using any dynamic procedure in turbulent and transitional channel flow. In the HPF Smagorinsky, model the mean shear is removed by applying a high-pass filter to the grid-filtered velocity field. The turbulent eddy-viscosity and strain rate is then computed from the high-pass filtered velocity field rather than from the grid-filtered velocity field.

Numerical simulation

ANSYS Fluent was used in this study for numerical simulation. The Reynolds numbers for the simulations were $Re=5, 10, 15, 20, 50, 100, 300 \times 10^4$ over a symmetric NACA airfoil 0025 at $0^\circ, 5^\circ$ and 10° is simulated here. The initial and boundary condition are defined based on standard room temperature and pressure.

Boundary conditions: The atmospheric far-field pressure with specified Mach number (calculated based on Reynolds number and the attack number) is used as boundary for all far field regions. The wall surface is taking no-sliding condition, and the initial condition of the flow field is setting in accordance with the free stream.⁸ The geometry shown in Figure 1 is discretized using a 3D structured mesh of more than 350,000 quadrilateral Finite volumes for the RANS models while 4 million were used for the LES. The domain extends from 12 chords lengths upstream, upper and lower heights to 21 chord lengths downstream. To generate an elliptical 3D C-type grid, the wingspan is 12 chords.^{3,8,14,22–24}

The convergence criteria selected for this problem is 10^{-5} for the scaled absolute residuals of continuity, 10^{-6} for the scaled absolute residuals of momentum equations and lastly 10^{-9} for the energy equation. For cases simulated with RANS, the convergence criterion

for turbulent viscosity was 10^{-5} . Also, the convergence of airfoil's lift and drag at the stationary result is considered.

Table 1 details the computational domain used including the number of cells in each direction, time step and quality of the mesh that has been used in this study. The boundary layer mesh has been also supplemented to divide the surface of the airfoil into two inner and outer regions. The inner region has a very fine mesh for E-LES over the entire domain. The governing equations are discretized over a

domain based on element volumes. It should be note that the mesh and time sensitivity analysis have been performed for different cases and accordingly the time the total simulation and time step was different for various cases to ensure the mesh and time independency for each case. Accordingly, the solution imbalances at the starting point was less than 1%, and the residual levels $3E-6$ is considered to converge in this study.

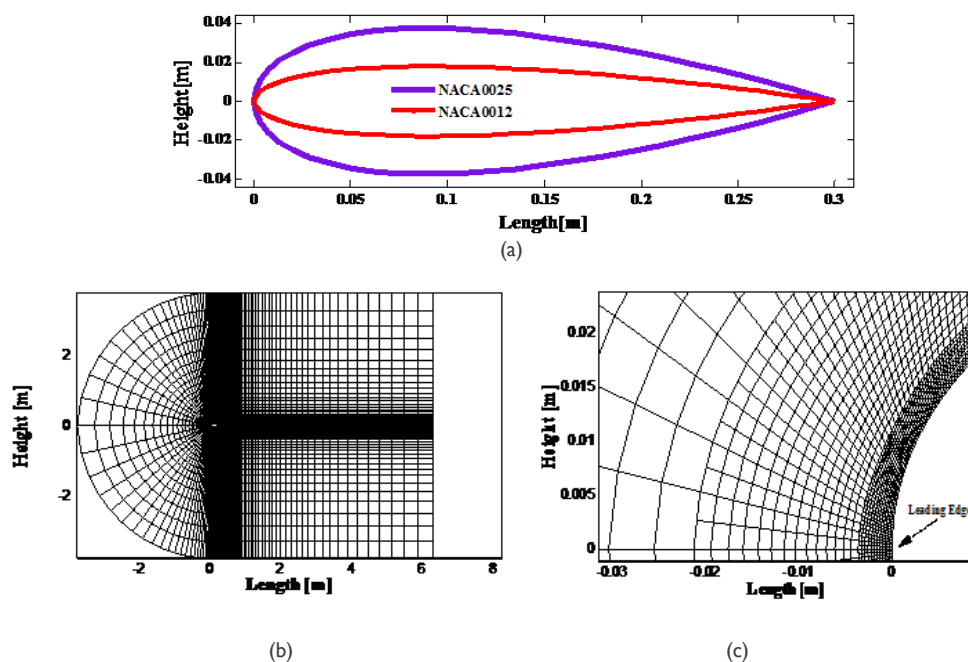


Figure 1 a) NACA Airfoils cross section, b) Computational domain for the numerical simulations at the midspan and c) Expanded domain close to leading edge.

Table 1 The detail grid information of the numerical domain for various turbulent approaches

Case	Turbulent approach	Number of cells along the x,y and z directions	Cells attached to the airfoil surface	Quality of the mesh	Time step [s]
1	RANS models	80x80x30	918	Coarse	2×10^{-1}
2	RANS models	100x100x36	3616	Fine	1×10^{-2}
3	LES	200x200x100	39876	Coarse	4×10^{-4}
4	LES	400x400x150	430870	Coarse	3×10^{-6}
5	E-LES	200x200x100	39876	Coarse	4×10^{-4}
7	E-LES	400x400x150	430870	Fine	3×10^{-6}

Numerical validation

For $Re > 150 \times 10^3$ the Spalart-Allmaras model predicts the flow separation fairly accurately. However, the E-LES approach is needed for lower Reynolds number. Consequently, the numerical results of the RANS model and E-LES approach were validated separately.

There are reliable experimental data^{10,25} for the NACA 0025 airfoil at $Re = 50, 100, 150$ and 200×10^3 at an angle of attack of 5° . The velocity profiles at different line around the airfoil has compared with the experimental results also and show 5 to 13% variation. The

measured pressure distributions on the upper surface of the NACA 0025 airfoil are compared with the results here.

Figure 2 is the pressure distributions on the upper surface of the NACA0025 airfoil when $\alpha = 5^\circ$ at various Reynolds number with good agreement between the numerical and experimental results. While E-LES was used for all other Reynolds numbers, for $Re = 200 \times 10^3$ the Spalart-Allmaras model is used and predicts the pressure distribution well. From Figure 2 there is the maximum deviation between the experimental and numerical results at the leading edge and the separation point.

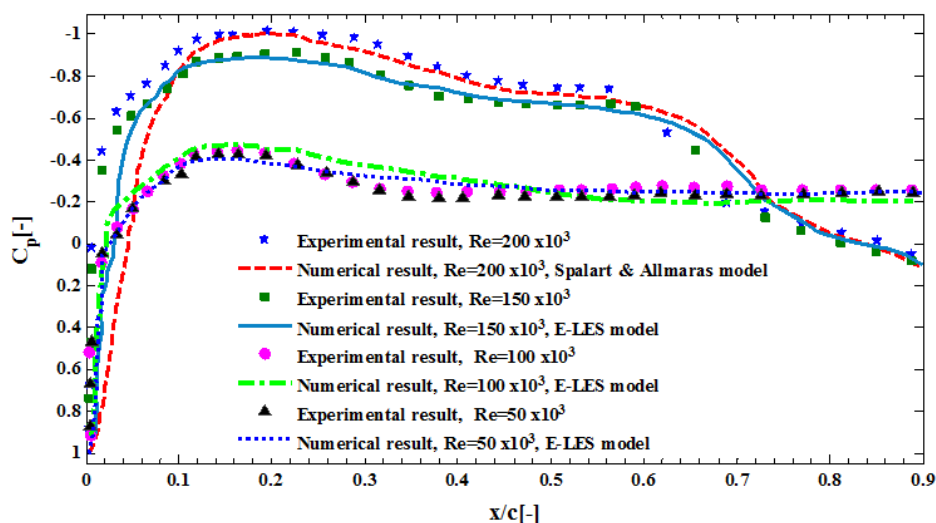


Figure 2 Pressure distributions on the upper surface of the NACA 0025 airfoil when $\alpha = 5^\circ$ at various Reynolds number.

Results and discussion

Comparison of various turbulent approaches with experimental results at low and high Reynolds number

Figure 3A shows flow visualization of a single smoke wire located upstream of the airfoil at $Re=50, 100$ and 150×10^3 . In the case of non-reattached flow, large-scale structures are shed alternatively from the upper and lower sides of the airfoil wake. Whereas, for reattached flow the bubble forms on the upper surface, similar periodicity is observed in the wake.

The Spalart and Allmaras model Figure 3B shows only a small bubble at the trailing edge for Reynolds numbers less than 100×10^3 . It must be added that the Spalart and Allmaras model is in fair agreement for Re higher than 150×10^3 ; although no agreement with experiments

was obtained at lower Re ($>100 \times 10^3$) based on Figure 3. The Spalart and Allmaras model and experimental pressure distributions on the upper surface of the NACA 0025 airfoil when $\alpha = 5^\circ$ at $Re=200 \times 10^3$ show a similar behavior that proves the application of this turbulent model for $Re=200 \times 10^3$. For $k-\varepsilon$, Figure 3C, no separation can be detected. Despite this, the average pressure distribution around the airfoil for both turbulent models is similar for both methods.

LES shows better agreement with the experimental results (Figure 3D). Although the flow field predicted by LES at $Re < 150 \times 10^3$ was closer to experiments with respect to the formation of bubble; it does not provide a good agreement with experiments above 10^5 . For example, bubble formation occurs only on the upper surface of the airfoil; whereas, according to the experiment at $Re=100 \times 10^3$ strong separation occurs at both surfaces of the airfoil.

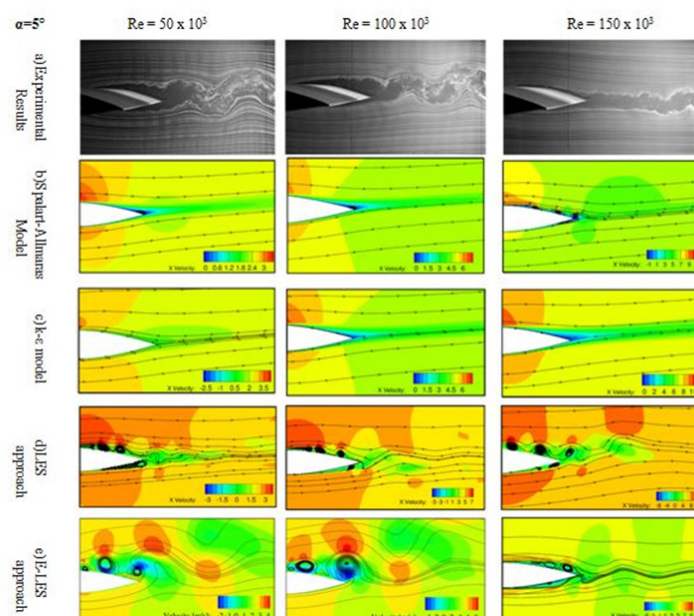


Figure 3 Experimental result using smoke-wire flow visualization based on Yarusevych³⁵ and Numerical results of wake and vortices at trailing edge and $\alpha = 5^\circ$, velocity along the x-direction [m/s].

While LES can accurately capture transition in the boundary layer of a NACA0025 airfoil at low Reynolds number ($10^3 \sim 10^4$) and provide further insight into the physics of the separation bubble,²⁶⁻²⁸ LES with a coarse mesh could not compute boundary layer separation precisely²⁰ because there needs to be a very small time step ($\sim 10^{-9}$) which is impossible with available resources. An alternative solution, which balances both cost and accuracy efficiency was E-LES. In this method, a very fine mesh may be used close to the airfoil boundary (inner region) where the flow is solved using LES; the outer region is calculated using a RANS model with a comparatively coarse mesh. As illustrated in Figure 3E, excellent agreement with experimental results are captured using E-LES for all experimental Reynolds numbers considered here. The Spalart-Allmaras turbulence model reasonably predicted the flow field at high Re numbers or high angle of attacks ($\alpha > 5^\circ$) (Figure 3) where the Spalart-Allmaras turbulence model predicted the formation of a bubble on the upper surface of the airfoil. On the other hand, the processing cost of the LES methods is prohibitively high for $Re > 100 \times 10^3$.

LES and E-LES showed accurate prediction of flow around the airfoils at low Re. On the other hand, the k- ϵ and Spalart-Allmaras models predict the time average pressure distribution in fair agreement with experimental results at high Reynolds numbers and angle of attacks. In addition, the Spalart-Allmaras model was able to capture

the bubble formation at the trailing edge as did LES. At $Re = 3 \times 10^6$, the boundary layer formed around the airfoil is fully turbulent³ and all turbulent models can reasonably predict the flow parameters. Consequently, LES approaches must be used for Re below critical Reynolds number ($Re_{cr} = 150 \times 10^3$) and for higher Re numbers, the Spalart-Allmaras model can predict the time average pressure distribution. E-LES was effective as it required about 10^{10} cells and a time step one to two orders of magnitude greater than LES's time step thanks to the coarser mesh. It should be noted that the value of y^+ for all of the cases were in the range of 0.7 to 1.45.

Three dimensional structure of the turbulent flow around NACA 0025 at $Re = 100 \times 10^3$ and $\alpha = 5^\circ$

Figure 4 are the velocity contour and iso-surfaces velocity along the three axes for NACA 0025 at $Re = 100 \times 10^3$ and $\alpha = 5^\circ$. As observed in Figure 2, the strongest wake vortex shedding occurs at $Re = 100 \times 10^3$, consequently, the 3D flow structure around the airfoil was analyzed. There is a clear 2-D velocity along the x and y directions. Iso-surface planes show no variation in the z-direction. Velocity fluctuations along the x-direction are evident under the trailing edge. Four bubbles can be seen in Figure 4B & 4C where the velocity along y-direction is plotted.

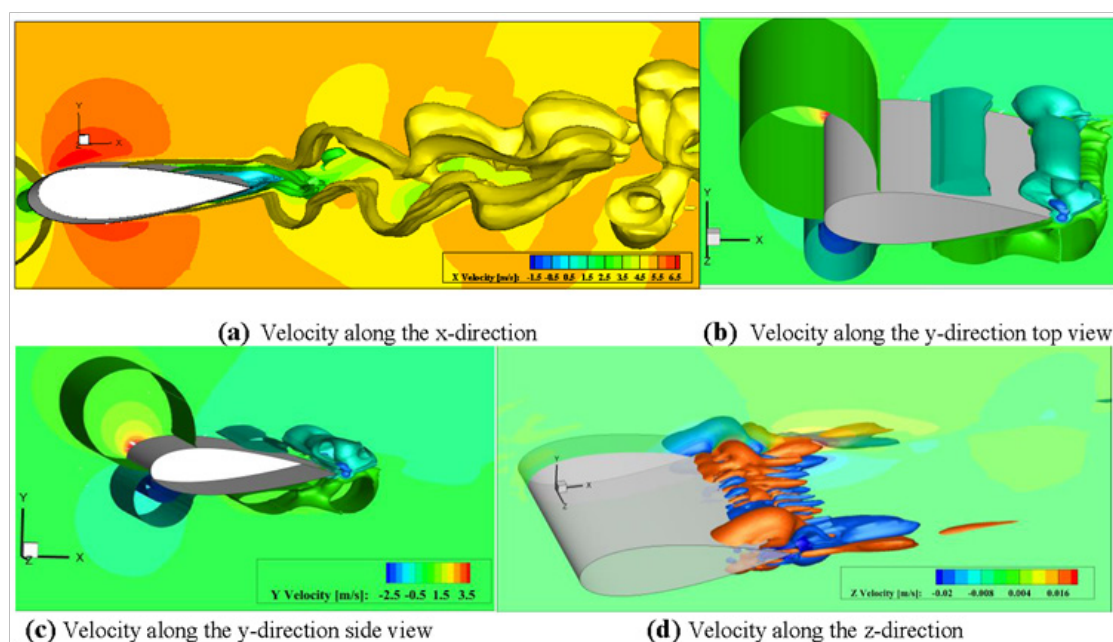


Figure 4 Velocity is o-surfaces along a) x-direction b) y-direction c) y-direction and d) z-directions for NACA 0025 at $Re = 100 \times 10^3$ and $\alpha = 5^\circ$.

Impact of reynolds number and angle of attack on boundary-layer separation

The reverse flow along the top surface of the airfoil was observed versus time at $x/c = 0.4$ and $Re = 100 \times 10^3$ (see Figure 3E and $Re = 100 \times 10^3$) with a roughly 5% overestimation of separation location compared with experimental results that indicated the separation at the maximum thickness location. Vortex structures are seen on the top surface of the airfoil when Re is less than $Re = 150 \times 10^3$. The flow reattached at $Re = 150 \times 10^3$, in agreement with the experiments (see Figure 3 and $Re = 100 \times 10^3$).

For Re numbers greater than 150×10^3 , a separation bubble is formed on the upper surface close to the trailing edge. As the Reynolds

number increases, the separation length decreases slightly, located between $x/c = 0.60$ and 0.94 . In contrast, decreasing the Reynolds number from 150×10^3 to 100×10^3 or 50×10^3 , brings about significant changes in the boundary layer development Figure 5A. The maximum thickness of the separation zone at the upper side of the airfoil is responsible for boundary-layer separation for $Re \sim 100 \times 10^3$. As shown in Figure 5B, maximum separation thickness occurs at 150×10^3 . Also, for Re numbers greater than 20×10^3 , a very thin (smaller than 3cm) separation zone is observed. Also, it can be seen that the most critical behavior occurs at Re around 100×10^3 , before this Re number, maximum thickness increases and after that this parameter decreases. The critical flow behavior at the same Reynolds number was observed experimentally.²⁹

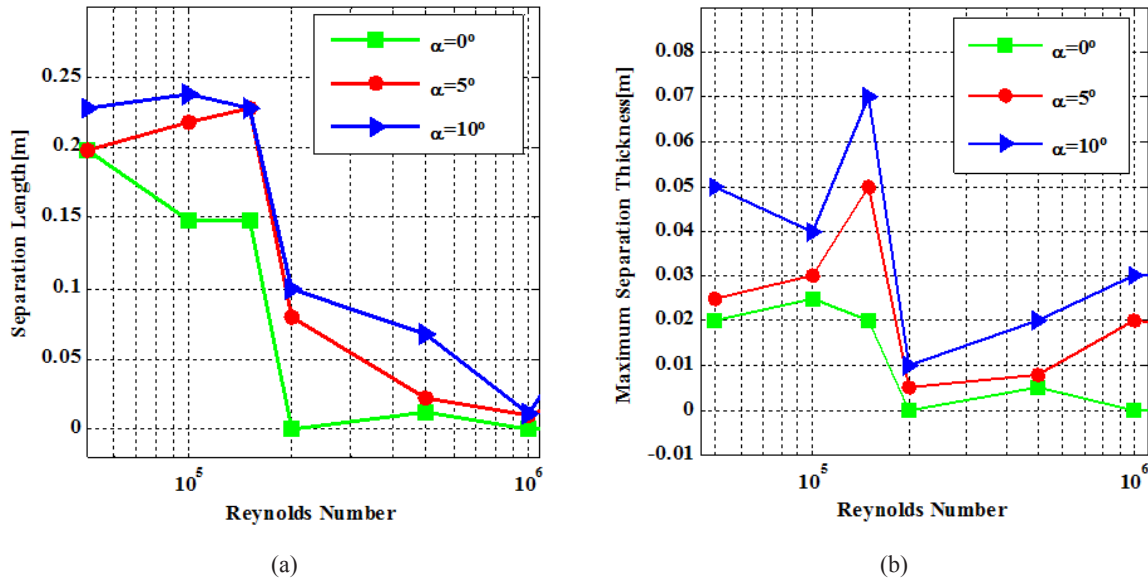


Figure 5 a) Length and b) Thickness of the Separation as function of Reynolds number at various angles of attack, NACA0025.

To study flow separation around the airfoils at low Reynolds number and close to dynamic stall, LES and E-LES must be used. Figure 6 shows the NACA 0025 airfoil boundary-layer separation at $Re = 100 \times 10^3$ for a range of angles of attack. Because of the thickness of this airfoil, separation is evident, even for zero angle of attack at $x/c = 0.48$. However, the separated flow reattaches to the airfoil top surface at $x/c = 0.96$. In contrast, reattachment at other angles of attack

that causes the vortex shedding at the trailing edge cannot be detected. In addition, the thin but lengthy separated layer is the main reason for the constant-pressure region between $x/c = 0.78$ and trailing edge.³⁰ As angle of attack increases, the separation point moves towards the leading edge¹⁴ until reaching 15% of the chord for $\alpha = 15^\circ$. For higher Reynolds number at $\alpha = 5^\circ$, the separated boundary-layer reattaches to the airfoil surface, forming a laminar separation bubble.

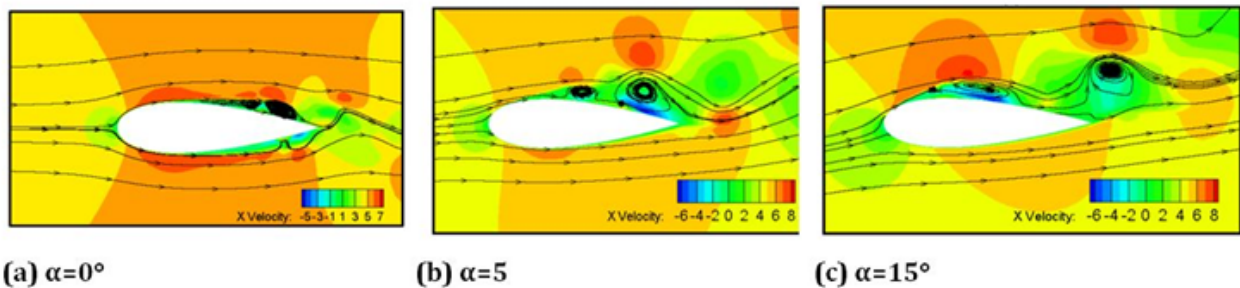


Figure 6 Impact of different angle of attacks on the boundary-layer separation at $Re = 100 \times 10^3$.

As seen in Figure 7A, the separation length over the NACA 0025 airfoil shows an approximate linear growth for increased angle of attack as well as for the maximum separation thickness (Figure 7B). In addition, it can be noted that a significant change in the flow field behaviour is detected comparing the first three lowest Re numbers and the three highest one in Figure 7A indicating two different regimes of the separation and the flow field.

Velocity fluctuation of flow close to upper boundary

Figure 8 shows the streamwise velocity during transition for $Re = 50 \times 10^3$, 100×10^3 , and 150×10^3 at different locations on top of the NACA 0025 airfoil. Initially, no significant fluctuations are observed in the laminar flow region until $x/c = 0.77$ at $Re = 50 \times 10^3$. It can be claimed that as angle of attack increases, increment in the magnitude of the velocity fluctuation and decrement of its frequency occurs. At

$Re = 50 \times 10^3$, the separated wake frequencies are roughly 27, 21 and 14 Hz for $\alpha = 0^\circ$, 5° and 10° respectively. For increased Reynolds number, the frequency of the streamwise velocity along x-direction increases. This observation can be seen in Figure 8B & 8C that these two diagrams represent the streamwise velocity at $\alpha = 5^\circ$. Beyond $x/c = 0.77$ Figure 8C, the fluctuations were random. The highest frequency was observed at Figure 8C that represent the fluctuation of the velocity at $x/c = 0.77$ when $Re = 150 \times 10^3$ (~110Hz).

Yarusevych et al.,¹⁰ have identified two distinct sets of data corresponding to the two flow regimes. They claimed that the fundamental frequency exhibits a power law dependency on the Reynolds number of the form $f_0 \sim Re^n$ with respect to these two flow regimes. Particularly, the value of the exponent n (the power of Re in $f_0 \sim Re^n$) for $Re < 150 \times 10^3$ varies from n obtained by curve fitting for $Re > 200 \times 10^3$. They applied the power-law to the

experimental data on airfoil profiles and the results are presented with solid lines in Figure 9 for both two regimes at various attack angles. As shown in the upper side of Figure 9, the values of the exponent n are collected from literature^{10,11,31,32} and used to find the best fits to numerical results. As it can be seen from the curve fits, the numerical data closely follow the power-law dependency established by experimental results.^{10,11,31,32} Nonetheless the second part of Figure 9 reports the percentage variations between the experimental and numerical results. The maximum and minimum variations are 14 and 1 percent for the flow at $Re=50 \times 10^3$ and $\alpha = 0^\circ$ and the flow at $Re=100 \times 10^3$ and $\alpha = 5^\circ$ respectively. In general, it can be seen that as Re increases the variation between the direct numerical simulation and the experimental data decreases. Finally, it can be claimed that the change of angle of attack does not have a pronounced impact on the behavior of the natural frequency.

The comparative analysis of the experimental results^{10,11,31,32} and the boundaries of the two flow regime proposed show a change of the flow regime leads to a change in the dependency of the fundamental frequency on the Reynolds number. It can be surveyed in more detail using the Strouhal number. Thus, the last parameter that is investigated in this study is a modified form of the Strouhal, the universal Strouhal number, $St^* = f_0 \frac{d^*}{U_0}$ where d^* represents the vertical distance between two vortices forming in the near-wake region and U_0 is the mean velocity of the flow field. The variation of the resulting Strouhal number with the Reynolds number is shown in Figure 10 for both experimental^{10,11,31,32} and numerical results obtained in present study. It can be seen that both experimental and numerical results show minor scatter between $St^*=0.13$ and 0.25 ; while, the average values for the numerical and experimental universal numerical Strouhal numbers are $St^* \sim 0.1789$ and 0.1718 , which are in good agreement with each other and also the Strouhal number reported for cylinder, ~ 0.2 .

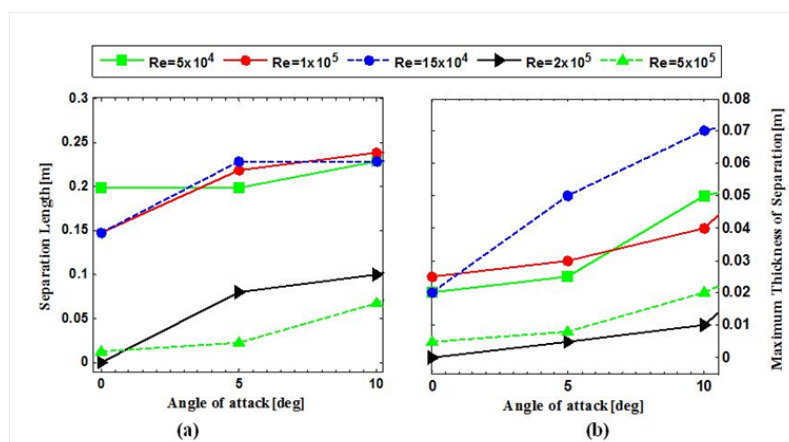
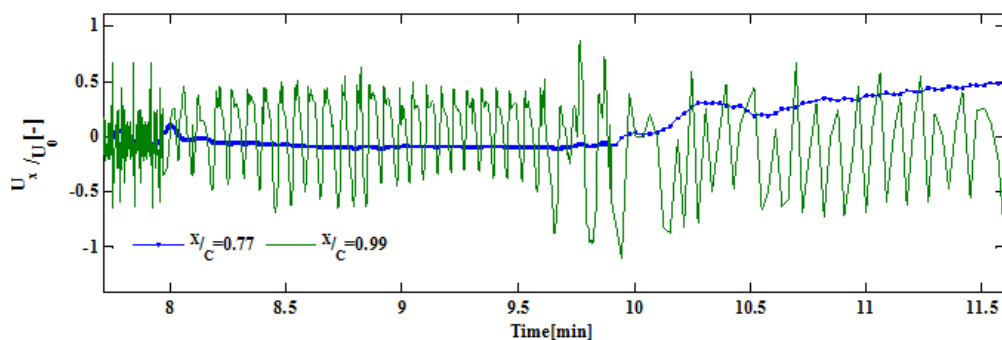
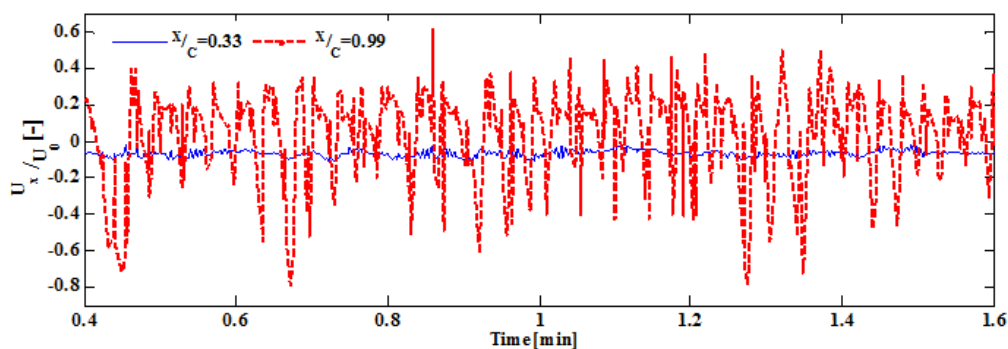


Figure 7 a) Separation length, and b) Maximum separation thickness as a function of angle of attack for NACA0025 and different Reynolds numbers.



a



b

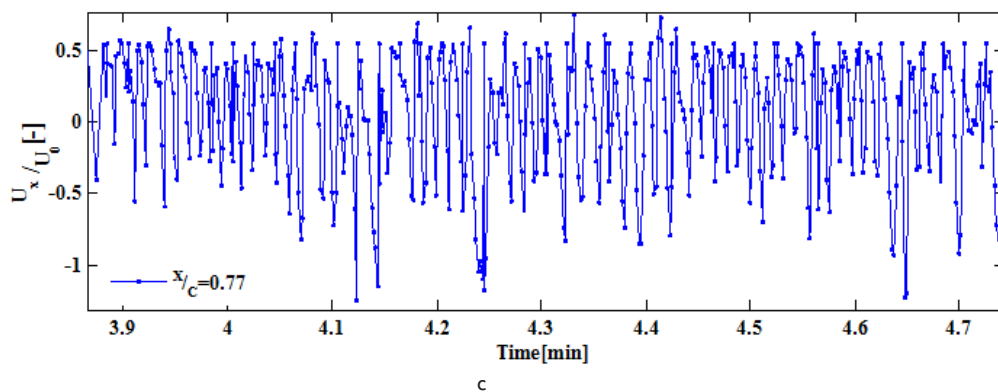


Figure 8 Stream wise fluctuating velocity component signals for NACA0025 at, a) Reynolds number 50×10^3 , $\alpha = 0^\circ, 5^\circ, 10^\circ$, b) Reynolds number 100×10^3 , $\alpha = 5^\circ$, and c) Reynolds number 150×10^3 , $\alpha = 5^\circ$

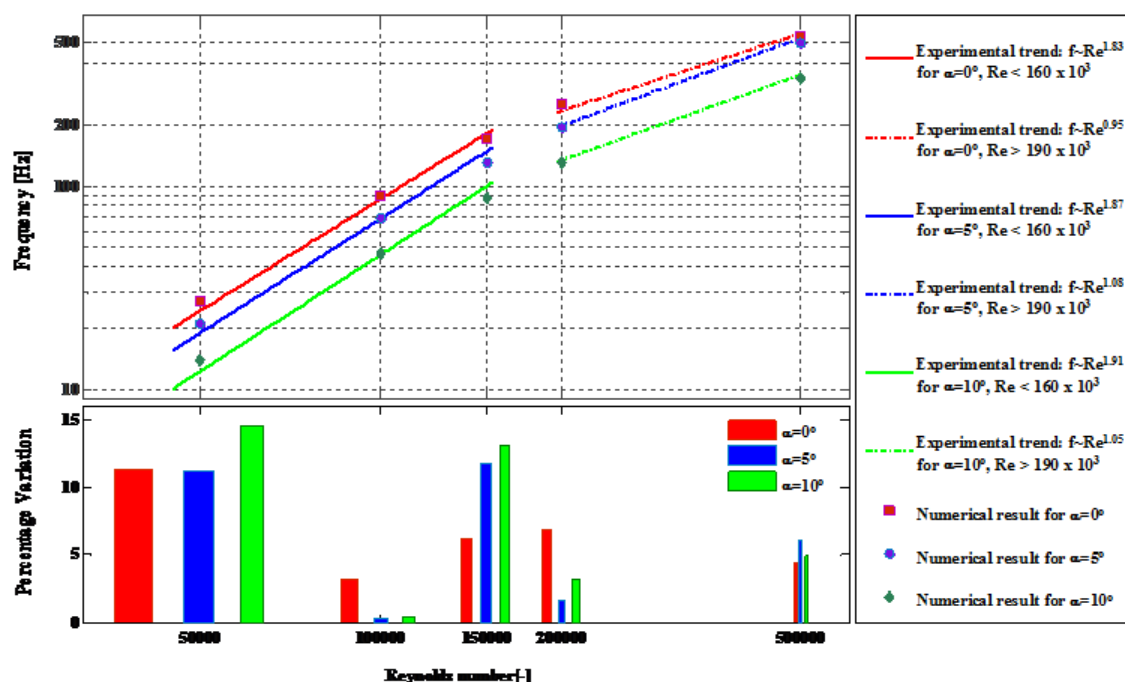


Figure 9 Experimental¹¹ and numerical variation of fundamental frequency f_0 with Reynolds number at $\alpha = 0^\circ, 5^\circ, 10^\circ$.

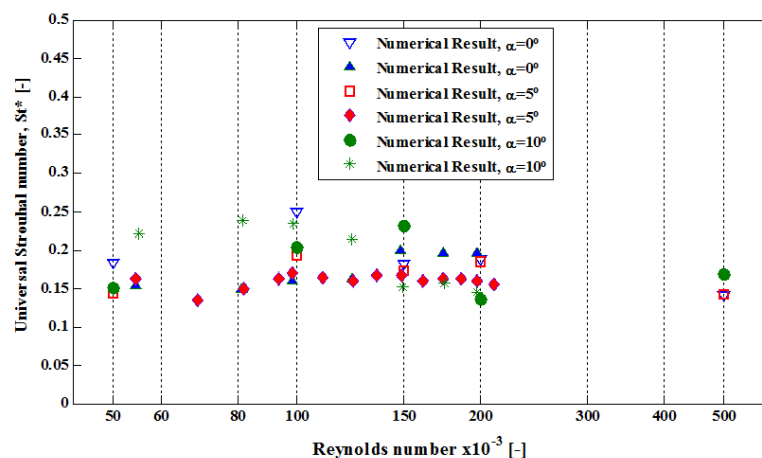


Figure 10 Universal vortex shedding frequency scaling based on wake flow geometry for experimental^{10,11,31,32} and numerical result obtained in this study.

Conclusion

Boundary layer and wake development on a NACA 0025 airfoil at low Reynolds numbers has been studied numerically. While different numerical approaches provide various results, LES and E-LES showed good agreement (especially E-LES). These methods only can be employed for low Re because of high computational cost for higher Re numbers. On the other side, the Spalart-Allmaras model captured the experimental data at higher Reynolds numbers or angles of attack in which LES approaches can be not used because of computational limits. The comparison between the numerical and experimental results suggests that for the LES approaches intensively fine mesh must be used to describe the boundary layer separation (even for low Reynolds numbers) while the RANS models cannot give accurate results.

The flow separation onset position and reattachment points on the airfoil contours have been calculated and compared on a NACA0025 airfoil at various angles of attack and Reynolds numbers. With LES and E-LES, separation at the zero angle of attack is captured. It can be seen that as Reynolds number increases, the airfoils demonstrate better aerodynamic performances based on this numerical analysis. Furthermore, it can be concluded that the aerodynamic performance of the airfoil increases as the separation point moves towards the trailing edge. Laminar boundary layer separation occurs on the upper surface of the NACA 0025 airfoil. While the flow reattaches to the airfoil surface at $Re=150 \times 10^3$, for $Re=100 \times 10^3$, the strongest wake shedding is observed in which separated shear layer fails to reattach to the airfoil surface.

Nomenclature

C_L	Normalized lift coefficient
C_D	Normalized drag coefficient
c	Airfoil chord [m]
d	Distance to the closest wall [m]
f_0	natural frequency [Hz]
G_k	Generation of turbulence kinetic energy due to the mean velocity gradients
G_b	Generation of turbulence kinetic energy due to buoyancy
ξ_t	$\text{Min}(0.1, \Delta U / \omega_t \Delta x_t)$
I	Unit tensor
M_t	Turbulent Mach number
$M_{w,i}$	The molecular weight of species
L_s	The separation length [m]
p, p_0	Local and free stream static pressure [Pa]
Re	Reynolds number $(U_0 c / \nu)$
T	The static temperature [K]
T_s	The thickness of separation [m]

Table Continues...

S	Magnitude of the vorticity [1/s]
S_m	Mass added to the continuous phase
U_0	Free stream velocity in x-direction [m/s]
u, v	x and y fluctuating velocity components [m/s]
x, y	Stream wise and vertical coordinates
α	Angle of attack [deg]
$\sigma_k, \sigma_\epsilon$	Turbulent Prandtl numbers for k and ϵ
μ	Viscosity [kg/ms]
ν	Kinematic viscosity of air [m ² /s]
ρ	Density of air [kg /m ³]
τ	Stress tensor [N/m ²]
ω_t	Wall vorticity at the trip [1/s]
ΔU	Velocity difference between field point and at the trip [m/s]
Δx_t	Grid spacing along the wall at the trip [m]
Y_i	The mass fraction of species
Y_M	Fluctuating dilatation contribution in compressible turbulence to the overall dissipation rate $(2\rho\epsilon M_t^2)$

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Conflicts of interest

Author declares that there is no conflicts of interest.

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